Notes:
1. This AD schedule is applicable to Airbus Helicopters Deutschland BO105 A, BO105 C, BO105 D, BO105 S, BO105 LS A-1, BO105 LS A-3 helicopters manufactured by Eurocopter Deutschland (formerly Messerschmitt-Bolkow-Blohm GmbH) under EASA Type Certificate No. R.011 (formerly LBA TCDS No. 3025).

2. The European Aviation Safety Agency (EASA) is the National Airworthiness Authority (NAA) responsible for the issue of State of Design Airworthiness Directives (ADs) for these helicopters. State of Design ADs can be obtained directly from the EASA website at http://ad.easa.europa.eu/

3. This AD Schedule includes Federal Aviation Administration (FAA) ADs mandated by EASA.

4. The date above indicates the amendment date of this schedule.

5. New or amended ADs are shown with an asterisk *

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The State of Design ADs listed below are available directly from the National Airworthiness Authority (NAA) websites. Links to NAA websites are available on the CAA website at https://www.aviation.govt.nz/aircraft/airworthiness/airworthiness-directives/links-to-state-of-design-airworthiness-directives/ If additional NZ ADs need to be issued when an unsafe condition is found to exist in an aircraft or aeronautical product in NZ, they will be added to the list below .............................................. 13

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## DCA/BO105/1B Airworthiness Directive Compliance at Initial Airworthiness Certificate Issue

### Applicability:

### Note 1:
DCA/BO105/1B revised to cancel LBA AD 2001-281 and LBA AD 1976-136/2, which have both been superseded by EASA AD 2021-0142. New or amended ADs are shown with an asterisk *

### Requirement:
Compliance with the following Luftfahrt-Bundesamt (LBA) Airworthiness Directives and EASA Airworthiness Directives (as applicable) is required:

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<tr>
<th>Luftfahrt- Bundesamt (LBA) / EASA / Transport Canada AD No:</th>
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<tr>
<td>LBA AD 1973-025</td>
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<tr>
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<tr>
<td>LBA AD 1975-106/2</td>
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<td>Hose Assemblies - Inspection and Replacement</td>
<td>Refer to LBA AD 1975-106/2</td>
</tr>
<tr>
<td>LBA AD 1975-171 (In German)</td>
<td>MBB BO105 SB 90-11 Rev 1 dated 18 Jul 1975</td>
<td>Engine Compartment Deck Socket Connection - Rework per MBB BO105 SB 90-11 Rev 1</td>
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<td>LBA AD 1975-172/2</td>
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<td>LBA AD 1975-238</td>
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<td>Right Outboard Engine Mount Flange - Flange Replacement</td>
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<tr>
<td>* LBA AD 1976-136/2</td>
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<td>Cancelled</td>
<td>Refer to EASA AD 2021-0142</td>
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<tr>
<td>LBA AD 1976-256/3 (In German)</td>
<td>MBB BO105 AB No. 13 rev 2 dated 17 Dec 1979</td>
<td>Rotor Brake - Inspect and modify per MBB BO105 AB No. 13</td>
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<td>LBA AD 1976-317/2</td>
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<td>LBA AD 1977-306</td>
<td>MBB BO105 SB 10-7 rev 1 dated 2 Nov 77 &amp; MBB BO105 SB 60-37 dated 27 Jun 77</td>
<td>Input Shaft and Ram Air Ventilation - Replacement and Modification</td>
<td>Refer to LBA AD 1977-306</td>
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<tr>
<td>Document</td>
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<td>Description</td>
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<tr>
<td>LBA AD 1979-136</td>
<td>MBB BO105 ASB 17 dated 20 Feb 1979</td>
<td>Exhaust Pipe Clamps - Inspection and Replacement</td>
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<tr>
<td>LBA AD 1979-137</td>
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<td>Drive Shaft Fairing Asbestos Cloth - Inspection</td>
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<tr>
<td>LBA AD 1979-444</td>
<td>MBB BO105 SB 60-46 dated 20 Aug 1979</td>
<td>Engine Cowling - Rework</td>
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<tr>
<td>LBA AD 1979-235/2</td>
<td>MBB BO105 SB 40-38 rev 1 dated 13 Aug 1979</td>
<td>Dual Hydraulic Boost System - Modification</td>
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<td>LBA AD 1978-295/2</td>
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<td>Tail Rotor Blades and Heads - Replacement - Terminating Action</td>
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<tr>
<td>LBA AD 1980-95/2 (In German)</td>
<td>MBB BO105 AB No. 19 rev 1 dated 27 March 1980</td>
<td>Bendix Drive Shafts - Inspect and modify per MBB BO105 AB No. 19</td>
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<tr>
<td>LBA AD 1980-208</td>
<td>MBB BO105 ASB AB No. 54 and 10-37 dated 27 June 1980</td>
<td>Main Gearbox - Rework</td>
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</tr>
<tr>
<td>LBA AD 1984-177/2</td>
<td>MBB BO105 SB 80-81 rev 1 dated 4 Dec 1984, MBB BO105 SB 80-4 rev 1 dated 27 Nov 1984 &amp; Siren SB 10A dated 13 Aug 1984</td>
<td>Cargo Hook - Accomplish SB Actions</td>
<td>Refer to LBA AD 1984-177/2</td>
</tr>
<tr>
<td>LBA AD 1984-128</td>
<td>MBB BO105 ASB 23 dated 26 Jul 1984</td>
<td>Tail Rotor Blades - Accomplish SB Actions</td>
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<tr>
<td>LBA AD 1983-124/2</td>
<td>MBB BO105 ASB 22 rev 1 dated 17 Apr 1984</td>
<td>Tandem Hydraulic Units - Spring Replacement - Terminating Action</td>
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<td>LBA AD 1985-50 (In German)</td>
<td>MBB BO105 AB No. 24 dated 18 February 1985</td>
<td>Tail Rotor Control Rods - Inspect and replace per MBB BO105 AB No. 24.</td>
<td>Refer to LBA AD 1985-50 (In German)</td>
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<td>LBA AD 1985-265</td>
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<td>Tandem Hydraulic System - Modification</td>
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<td>LBA AD 1986-106</td>
<td>MBB BO105 ASB 28 dated 27 Mar 1986</td>
<td>TR Gearbox Attach Bolts - Replacement</td>
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<td>LBA AD 1989-123/2 (In German)</td>
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<td>Dual Hydraulic System - Inspect and rework per MBB BO105 ASB No. 105-40-102</td>
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<tr>
<td>LBA AD 1991-204</td>
<td>MBB BO105 ASB BO105-60-105 dated 18 Nov 1991</td>
<td>PC Air Tube &amp; Pipe Assembly Clearance - Rework and Replacement</td>
<td>Refer to LBA AD 1991-204</td>
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<tr>
<td>AD Number</td>
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<td>Manufacturer</td>
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<tr>
<td>LBA AD 1995-458</td>
<td>Voltage Controller - Replacement</td>
<td>Eurocopter ASB 105-80-118 rev 1</td>
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<tr>
<td>LBA AD 1999-300/3</td>
<td>Main Rotor System Tension Torsion Strap - Inspection and Replacement</td>
<td>Eurocopter ASB 105-10-114 rev 2</td>
<td>Issued 31 Aug 1999</td>
</tr>
<tr>
<td>LBA AD 1999-289/3</td>
<td>Main Rotor System Tension Torsion Strap - Inspection and Replacement</td>
<td>Eurocopter ASB BO 105-10-113 rev 3</td>
<td>Issued 10 Nov 2000</td>
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<td>LBA AD 2001-281</td>
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<tr>
<td>EASA AD 2010-0049</td>
<td>Cyclic Stick Locking Device - Modification</td>
<td>Eurocopter ASB BO105-40-106</td>
<td>Issued 19 Dec 2008</td>
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<tr>
<td>LBA AD 2009-110R1</td>
<td>Rotor Control – Cyclic-Stick Locking Device – Modification</td>
<td>Eurocopter Canada Limited ASB-BO105 LS 40-10</td>
<td>Issued 08.05.2009</td>
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<td>LBA AD 2010-181</td>
<td>Engine – Power Turbine Speed – Operation Limitation</td>
<td>Eurocopter Deutschland GmbH ASB BO105-60-110 Rev. 1</td>
<td>Issued 03.03.2010</td>
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<td>LBA AD 2010-207</td>
<td>Time Limits / Maintenance Checks – Main Rotor Blades with Bolted Lead Inner Weight - Life Limitation</td>
<td>Eurocopter Deutschland ASB BO105-10-121 Rev. 1 MM BO105, Chapter 101-15 Airworthiness Limitations, Revision 26</td>
<td>Issued 18.10.2010</td>
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<tr>
<td>CF-2009-12</td>
<td>Eurocopter Canada ASB BO 105 LS 30-12, dated 12 December 2008, or later revisions approved by Transport Canada.</td>
<td>Tail Rotor Balance Weights Corrosion</td>
<td>BO 105 LS A-3</td>
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<tr>
<td>CF-2008-17R1</td>
<td>Eurocopter Canada ASB BO 105 LS-10-10 Revision 1, dated 8 January 2008, or later revision approved by Transport Canada.</td>
<td>Tension Torsion Straps</td>
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</table>

Note 2: Each part of this AD (each individual LBA AD, EASA AD and Transport Canada AD) shall be certified in the aircraft log book separately.

Note 3: Manufacturer service information at later approved revisions is acceptable to comply with the requirements of this AD.

Compliance: Initial compliance required before the issue of a New Zealand Certificate of Airworthiness, or at the next Review of Airworthiness (RA), or at the next annual inspection, whichever is the sooner, unless previously accomplished. Repetitive inspections, if required, are to be accomplished at intervals not to exceed the times specified in the individual LBA ADs, the EASA ADs and the FAA ADs, as applicable.

Effective Date: DCA/BO105/1 - 29 July 2010
DCA/BO105/1A - 27 September 2012
DCA/BO105/1B - 1 July 2021

DCA/BO105/2 Tail Rotor Blades – Inspection and Rework

Applicability: All model BO 105 series aircraft fitted with tail rotor blades P/N 105-31742, S/N all through 548.

Note: This AD pertains to fiberglass plastic loops embodied with LTA 74-64 per MBB SB 30-5 dated 29 March 1974 or later approved revisions.

Requirement: To prevent Tail Rotor Blade (TRB) failure, accomplish the following:
1. Secure TRB end caps per SB 30-8 dated 5 July 1974 or later approved revisions.
2. Accomplish the instructions in action 2.a) or 2.b) of LBA AD 1974-162 as applicable.

(LBA AD 1974-162 refers)

2. At every post flight inspection until the requirements in MBB SB 30-8 have been accomplished.

Effective Date: 29 July 2010
DCA/BO105/3 Main Rotor Head Nuts and Bolts – Inspection and Replacement

Applicability: All model BO 105 series aircraft fitted with quadruple nuts and Bendix tie bar retaining bolts.

Requirement: To prevent MRH failure, accomplish the following:

- Remove quadruple nuts P/N 105-14101.19 and 105-14101.20, and bolts P/N 105-14101.22 and 105-14101.23 and magnaflux inspect them for cracks. Replace cracked parts before further flight.

Note: MBB SB No. 10-18 dated 18 July 1975 and later approved revisions pertains to the subject of this AD.

Compliance: Within the next 100 hours TIS unless previously accomplished and thereafter at intervals not to exceed 300 hours TIS until the life limitation of 2400 hours TTIS.

Effective Date: 29 July 2010

DCA/BO105/4 Main Rotor Gearbox Oil Pump – Inspection and Rework

Applicability: All model BO 105 series aircraft fitted with a main rotor gearbox oil pump P/N 7631 955 112 with S/N all through 229, or oil pump P/N 7631 955 115 with S/N all through 229.

Requirement: To prevent failure of the main rotor gearbox sun and planetary gear carrier, accomplish the following:

1. Inspect the main gearbox chip detector per the instructions in chapter 10-1-2, page 17 of the BO 105 Maintenance and Overhaul Manual. If any chips are found accomplish corrective actions before further flight.

2. For oil pump P/N 7631 955 112:
   Replace the oil pump plunger and compression spring per the instructions in MBB SB No. 10-20 dated 25 May 1976 and MBB SB No. 10-16 dated 28 July 1975 or later approved revisions.

3. For oil pump P/N 7631 955 115:
   Replace the oil pump compression spring per the instructions in MBB SB No. 10-16.

Compliance: 1. Within the next 10 hours TIS unless previously accomplished and thereafter at intervals not to exceed 10 hours TIS until accomplishment of requirements 2 or 3, as applicable.

2. Within the next 100 hours TIS unless previously accomplished.

3. Within the next 100 hours TIS unless previously accomplished.

Effective Date: 29 July 2010
DCA/BO105/5  Tail Rotor Blade Grips – Inspection and Replacement  

Applicability: All model BO 105 series aircraft fitted with blade grips P/N 105-31711 and P/N 105-31722.

Requirement: To prevent failure of the tail rotor blades due to possible cracks in the inboard end and/or cracks in the laminated pack retaining bolt bore, accomplish the following:

1. Inspect the inboard end of the blade grips for cracks per the instructions in paragraph 2.A of MBB BO 105 SB No. 30-24 dated 1 December 1978 or later approved revisions. If any defects are found accomplish corrective actions before further flight.

2. Dye penetrant inspect the laminated pack retaining bolt bore per the instructions in paragraph 2.A of MBB BO 105 SB No. 30-24. If any defects are found accomplish corrective actions before further flight.

(LBA AD 1978-339 refers)

Compliance: 1. Within the next 100 hours TIS unless previously accomplished and thereafter inspect per the Special Inspection Instructions in the BO 105 Maintenance and Overhaul Manual at intervals not to exceed 100 hours TIS.

2. Within the next 100 hours TIS unless previously accomplished and thereafter inspect per the Special Inspection Instructions in the BO 105 Maintenance and Overhaul Manual at intervals not to exceed 600 hours TIS.

Effective Date: 29 July 2010

DCA/BO105/6  Tail Rotor Blade Grips – Inspection and Replacement  

Applicability: All model BO 105 series aircraft, all S/N fitted with tail rotor blade grips P/N 105-31711 and P/N 105-31722.

Requirement: To prevent failure of the tail rotor blades due to possible cracks in the clevis area of the blade grips, accomplish the instructions in MBB BO 105 AB No. 18 dated 15 March 1979 or later approved revisions. If any defects are found accomplish corrective actions before further flight.

(LBA AD 1979-144 refers)

Compliance: Within the next 10 hours TIS unless previously accomplished and thereafter at intervals not to exceed 100 hours TIS.

Effective Date: 29 July 2010

DCA/BO105/7  Bendix Shaft and Clutches – Inspection and Replacement  

Applicability: All model BO 105 series aircraft, all S/N fitted with Bendix shafts and Bendix clutches P/N 19E 143-1, 19E 144-1/2/3, 19E 145-1/2/3, 19E 146-1, 105 31503, 105 31504, 105 31505 or 105 31506.

Requirement: To prevent failure of the shaft and clutch due to possible corrosion of the inner sides of the diaphragms, inspect per the instructions in MBB SB No. 60-45 dated 20 July 1979 or later approved revisions. If any defects are found accomplish corrective actions before further flight.

(LBA AD 1979-443 refers)

Compliance: Within the next 100 hours TIS unless previously accomplished and thereafter at intervals not to exceed 1200 hours TIS or 2 years whichever occurs sooner.

Effective Date: 29 July 2010
DCA/BO105/8  Hoist Cable – Inspection and Replacement

Applicability: Model BO 105 series aircraft, all S/N fitted with rescue hoist P/N 0134-0353, 0136-1175 or 105-81101.

Requirement: To prevent failure of the hoist cable due to possible previous damage, accomplish the instructions in MBB AB No. 21 dated 14 September 1982 or later approved revisions. If any defects are found accomplish corrective actions before further flight.

(LBA AD 1982-099 refers)

Compliance: Before further hoist operation unless previously accomplished and thereafter after every hoist operation and for frequent actuation during a hoist mission after every 10th hoist cycle.

Effective Date: 29 July 2010

DCA/BO105/9  Cancelled - EASA AD 2018-0056 refers

Effective Date: 26 March 2018

DCA/BO105/10 Tail Rotor Balance Weights and Control Lever – Inspection and Replacement


Requirement: To prevent failure of the tail rotor balance weights due to possible corrosion, accomplish the following:

Inspect the tail rotor balance weights and control levers per the instructions in Eurocopter Deutschland (ECD) BO105 ASB No. ASB BO105-30-116 dated 23 September 2008 or later approved revisions.

If any defects are found which exceeds the acceptable limits specified in ASB No. ASB BO105-30-116, replace defective parts before further flight.

Note: The replacement of defective parts is not a terminating action for the repetitive inspection requirements of this AD.

(EASA AD 2008-0206 refers)

Compliance: By 29 August 2010 or the next 100 hours TIS whichever occurs sooner unless previously accomplished, and thereafter at intervals not to exceed 600 hours TIS or 48 months whichever occurs sooner.

Effective Date: 29 July 2010
DCA/BO105/11 Power Turbine Speed – Operational Limitation, Placard and AFM Amendment

Applicability: All model BO 105C, BO 105D and BO 105S aircraft series fitted with Rolls Royce Corporation (formerly Allison, Detroit Diesel Allison) 250-C20 series engines, except those aircraft embodied with ECD VTOL retrofit kit 105-80037.

Requirement: To prevent 3rd stage turbine wheel failure due to detrimental vibrations in a particular RPM range which could result in loss of engine power, accomplish the following:

1. Review the aircraft records or inspect both engines and determine if a 3rd stage turbine wheel P/N 23065833 is fitted to any of the aircraft engines.

   If a 3rd stage turbine wheel P/N 23065833 is found fitted, accomplish requirement 2 of this AD.

2. Install a placard next to the RPM indicator on the instrument panel in full view of the pilot per the instructions in Eurocopter Deutschland GmbH ASB BO105-60-110 revision 1, dated 3 March 2010 or later approved revisions.

   Min. Continuous 98%
   Min. Transient 95%
   ASB BO105-60-110

3. Amend the AFM and insert the applicable revision listed in table 1 into the aircraft AFM.

Table 1 – Model BO 105 AFM Revisions

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<th>Model Variant</th>
<th>AFM Revision</th>
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<td>BO 105C</td>
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<tr>
<td>BO 105S</td>
<td>CBS (Europe), CBS-2 (USA) and CDN-BS (Canada)</td>
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<tr>
<td>BO 105D</td>
<td>DB and DBS (Europe)</td>
<td>No. 4</td>
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</tbody>
</table>

Note 1: Rolls Royce Corporation 3rd stage turbine wheels P/N 23065833 are no longer in production.

Note 2: The placard on the aircraft instrument panel can be removed once affected 3rd stage turbine wheels P/N 23065833 are removed from service and both engines have turbine wheel P/N fitted which are not affected by this AD per the instructions in paragraph 2.C.(1)(d) of Rolls-Royce Corporation 250-C20 Series CEB A-1400 revision 3 dated 19 January 2009, or later approved revisions.

(EASA AD 2010-0128 refers)

             2. By 29 August 2010.

Effective Date: 29 July 2010
* DCA/BO105/12  EASA AD 2010-0153 Cancelled – EASA AD 2021-0142 refers

Effective Date:  1 July 2021

DCA/BO105/13 MRB Erosion Protective Shell – Inspection and Replacement


Requirement:  To prevent loss of the MRB erosion protective shell due to possible debonding which can result in an unbalanced main rotor, high vibration, damage to the aircraft and loss of aircraft control, accomplish the following:

1. Review the aircraft records or inspect the aircraft and determine the P/N of the main rotor blades and determine whether any blades have been replaced between September 2006 and March 2010. Affected main rotor blade P/N are listed in ECD ASB BO105-10-124 revision 1 dated 18 October 2010 or later EASA approved revisions.

   If an affected main rotor blade is found fitted, accomplish a one-time inspection of the erosion protective shell for debonding per the instructions in ASB BO105-10-124.

   If any debonding is found, replace the erosion protective shell at the time indicated in, and per the instructions in chapter 14-2 of the applicable BO105 Maintenance Manual (MM).

2. Affected main rotor blades listed in ASB BO105-10-124 fitted with erosion protective shells which have been replaced between September 2006 and March 2010 shall not be fitted to any aircraft unless the blades have been inspected and reworked as applicable, per the requirements in this AD.

Note:  Inspections and corrective actions accomplished prior to the effective date of this AD per the original issue of ECD ASB BO105-10-124 are acceptable to comply with the initial requirements of this AD. After the effective date of this AD, inspections and corrective actions must be accomplished per ASB BO105-10-124 revision 1 or later EASA approved revisions.

   (EASA AD 2010-0216-E)

Compliance:  1. Within the next 10 hours TIS, or next 4 flight cycles, or by 3 December 2010 whichever occurs sooner, unless previously accomplished.

2. From 3 November 2010.

Effective Date:  3 November 2010

DCA/BO105/14 Cancelled – DCA/BO105/15A refers

Effective Date:  27 September 2012

DCA/BO105/15A  Cancelled – EASA AD 2011-0091R2 refers

Effective Date:  8 November 2012
DCA/BO105/16 Emergency Float Kit – Inspection and Modification


Requirement: To prevent an unsafe condition accomplish the inspections and corrective actions specified in FAA AD 2011-25-01.

Compliance: By 26 July 2012

Effective Date: 26 January 2012

DCA/BO105/17 Tail Rotor Pitch Links – Inspection and Replacement


Requirement: To prevent failure of the tail rotor pitch link due to possible spherical bearing migration out of the bearing bore which could result in loss of aircraft control, accomplish the requirements in FAA AD 2012-13-11.

Compliance: At the compliance times specified in FAA AD 2012-13-11.

Effective Date: 27 September 2012
The State of Design ADs listed below are available directly from the National Airworthiness Authority (NAA) websites. Links to NAA websites are available on the CAA website at https://www.aviation.govt.nz/aircraft/airworthiness/airworthiness-directives/links-to-state-of-design-airworthiness-directives/

If additional NZ ADs need to be issued when an unsafe condition is found to exist in an aircraft or aeronautical product in NZ, they will be added to the list below.

2011-0091R2  Cancelled – EASA AD 2014-0230 refers  
Effective Date:  4 November 2014

* 2013-0015  Cancelled – EASA AD 2021-0142 refers  
Effective Date:  1 July 2021

2014-0180R1  Cancelled – EASA AD 2015-0166 refers  
Effective Date:  25 August 2015

2014-0230  Main Gearbox – Inspection  
Note:  The repetitive inspections of the MGB magnetic plugs for chips mandated by requirement (1) of EASA AD 2014-0230 per the instructions in ASB BO105-10-125 revision 3 and the BO105 MM may be accomplished by adding the inspection requirements to the tech log. The visual inspection may be performed and certified under the provision in Part 43 Appendix A.1 (7) by the holder of a current pilot licence, if that person is rated on the aircraft, appropriately trained and authorised (Part 43, Subpart B refers), and the maintenance is recorded and certified as required by Part 43.

If any deposits are found on the MGB magnetic plug per Changed Table 11-2 of ASB BO105-10-125 revision 3 during any inspections, then a maintenance engineer must accomplish all required corrective actions per ASB BO105-10-125 revision 3 before further flight.

Effective Date:  EASA AD 2014-0230  -  4 November 2014  
EASA AD 2014-0230 (note added)  -  26 November 2015

2014-0235  Cancelled – EASA AD 2015-0017 refers  
Effective Date:  10 February 2015

2015-0017  Rescue Hoist Control Grip – Inspection  
Effective Date:  10 February 2015

* 2015-0042  Cancelled – EASA AD 2021-0142 refers  
Effective Date:  1 July 2021

2016-0166  Cancelled – EASA AD 2016-0060 refers  
Effective Date:  6 April 2016

2015-0220  Electrical Power System – Inspection  
Effective Date:  23 November 2015

2016-0060  Cancelled – EASA AD 2016-0142 refers  
Effective Date:  2 August 2016
2016-0118-E MRB Erosion Protection Shell – Inspection
Effective Date: 21 June 2016

2016-0142R1 Swashplate Assembly – Inspection
Applicability: BO105 A, BO105 C, BO105 D, BO105 S and BO105 LS A-3 helicopters, all variants, all S/N, and BO105 LS A-3 helicopters, modified in accordance with EASA STC 10039633, or previously by LBA STC EMZ NR. 0654/3058 (so called “Superlifter”).
Effective Date: 2016-0142 - 2 August 2016
2016-0142R1 - 26 April 2018

2018-0056 Main Rotor Mast - Inspection
Effective Date: 26 March 2018

* 2019-0024 Cancelled - EASA AD 2021-0142 refers
Effective Date: 1 July 2021

* 2021-0142 Airworthiness Limitations
Applicability: BO105 A, BO105 C, BO105 D, BO105 S, BO105 LS A-1 and BO105 LS A-3 helicopters, all variants, all S/N, including BO105 LS A-3 helicopters modified in accordance with EASA Supplemental Type Certificate (STC) 10039633, or previously LBA Germany STC EMZ NR. 0654/3058 (commercially known as “Superlifter”).
Effective Date: 1 July 2021